Preliminary Amendment to Reissue Patent Application

AMENDMENTS TO THE CLAIMS

Amendments to the claims have been made in accordance with 37 C.F.R. 1.173 (b)(2) and appear as follows:

Claims 1-8 that appear in the issued patent remain pending.

9. A momentum control device for a spacecraft comprising:

a unitary structure;

a plurality of control moment gyros operatively connected to the unitary structure, each of the plurality of control moment gyros adapted to rotate about an axis to thereby generate a torque;

a series of mounting members disposed on the unitary structure and adapted to operatively support the plurality of control moment gyros;

a control system adapted to control the operation of each of the plurality of control moment gyros to thereby generate a desired resultant torque; and

an attachment device connected to the unitary structure and including a plurality of struts, each of the plurality of struts having a predetermined static stiffness characteristic and a predetermined passive damping characteristic, the plurality of struts adapted to transmit the desired resultant torque generated by the plurality of control moment gyros and to attenuate a transmission of vibration generated by the plurality of control moment gyros.

10. The device of claim 9, wherein the device includes between three and eight control moment gyros.

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- 11. The device of claim 9, further including a series of force sensors adapted to generate force information indicative of a force transmitted through each of the plurality of struts.
- 12. The device of claim 9, wherein the unitary structure forms a rigid arrangement that connects the series of mounting members and that has a stiffness characteristic that is greater than a stiffness characteristic of the attachment device.
- 13. The device of claim 12, wherein the unitary structure includes a plurality of terminal members and a plurality of elongated joining members.
- 14. The device of claim 9, wherein each of the plurality of struts includes a flex pivot.
- 15. A momentum control device for positioning a spacecraft comprising:

 a plurality of bodies, each of the plurality of bodies adapted to rotate about an axis to thereby generate a torque;

a series of mounting members adapted to rotatably support the plurality of bodies;

a unitary structure adapted to form a rigid arrangement having a predetermined stiffness

characteristic that connects the series of mounting members;

a control system adapted to control the operation of each of the plurality of bodies to thereby generate a desired resultant torque; and

an attachment device connected to the unitary structure and including a plurality of struts, each of the plurality of struts having a predetermined static stiffness characteristic and a predetermined passive damping characteristic, the plurality of struts adapted to transmit the

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desired resultant torque generated by the plurality of control moment gyros and to attenuate a transmission of vibration generated by the plurality of control moment gyros, wherein the stiffness characteristic of the unitary structure is greater than a stiffness characteristic of the attachment device.

- 16. The device of claim 15, wherein each of the plurality of bodies is a control moment gyro.
- 17. The device of claim 15, further including a series of force sensors adapted to generate force information indicative of a force transmitted through each of the plurality of struts.
- 18. The device of claim 15, wherein the unitary structure includes a plurality of terminal members and a plurality of elongated joining members.
- 19. The device of claim 15, wherein each of the plurality of struts includes a flex pivot.